AIAA CFD Drag Prediction Workshop
June 2001, Anaheim, CA

Test Conditions

For all cases, Reynolds No. = $3x10^6$ (Based on c_{ref} = 141.2 mm)

- Required Cases

1. Single Point
   Mach = 0.75
   Lift Coefficient = .500 ± .001
   Run on standard grid.

2. Drag Polar
   Mach = 0.75
   Angle of Attack (Deg) = -3°, -2°, -1°, 0°, 1°, 2°
   Generation of suitable grid is encouraged.

- Optional Cases

3. Constant C_L Mach Sweep
   Mach = .50, .60, .70, .75, .76, .77, .78, .80
   Lift Coefficient = .500 ± .001
   (Approx $\alpha \approx$ .86°, .56°, .13°, -.17°, -.22°, -.29°, -.37°, -.43°
   Note: These values included for guidance only!)
   Generation of suitable grid is encouraged.

4. Mach/C_L Matrix (Drag Rise Curve)
   Mach = .50, .60, .70, .75, .76, .77, .78, .80
   Lift Coefficient = .400, .500, .600 ± .001
   Generation of suitable grid is encouraged.

Notes:

- Simulations are to be "free air"; no wind tunnel walls or model suport systems are to be included.
- Boundary layer is to be fully turbulent, do not use the transition pattern specified in Agard 303.
- For the optional cases, interpolated forces and moments are acceptable (as is typically done for wind tunnel data reduction).
- Reference Geometry:
  S_{ref} = 0.1454 m^2 (full model), c_{ref} = 141.2 mm
  Moment Reference Center: x = 504.9 mm, z = 0